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 airfoil shapes for aircraft wings  
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 shape of the NACA airfoils is described  
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 The NACA airfoils are airfoil shapes for  
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 (NACA). The shape of the NACA airfoils is  
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were generated using analytical equations that describe the camber (curvature) of the mean-line (geometric centerline) of the airfoil section as well as the section's thickness distribution along the length of the airfoil. Later The NACA airfoil series - Stanford University The NACA airfoil section is created from a camber line and a thickness distribution plotted perpendicular to the camber line. The equation for the camber line is split into sections either side of the point of maximum camber position (P). In order to calculate the position of the final airfoil envelope later the gradient of the camber line is also required. The equations are: NACA 4 digit airfoil generator (NACA 2412 AIRFOIL) Airfoil database search (NACA 6 series) Search

the 1638 airfoils available in the databases filtering by name, thickness and camber. Click on an airfoil image to display a larger preview picture. There are links to the original airfoil source and dat file and the details page with polar diagrams for a range of Reynolds numbers. NACA 6 series Airfoil database search Download Ebook The Naca Airfoil Series Clarkson University The Naca Airfoil Series Clarkson University FeedBooks provides you with public domain books that feature popular classic novels by famous authors like, Agatha Christie, and Arthur Conan Doyle. The site allows you to download texts almost in all major formats such as, EPUB, MOBI and PDF. The Naca Airfoil Series Clarkson University National Advisory Committee for Aeronautics

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Theoretical data are presented for NACA 6a-series basic thickness forms having the position of minimum pressure of 30, 40, and 50 percent chord and with thickness ratios varying from 6 percent to 15 percent.

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Aeronautics airfoils. During the late 1920s and into the 1930s, the NACA developed a series of thoroughly tested airfoils and devised a numerical designation for each airfoil — a four digit number that represented the airfoil section's critical geometric properties.

*NACA airfoil - Wikipedia*

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## Highlighting Flow Separation with Flat Plate Design

The NACA 6a-series airfoil sections were designed to eliminate the trailing-edge cusp which is characteristic of the NACA 6a-series sections. Theoretical data are presented for NACA 6a-series basic thickness forms having the position of minimum pressure of 30, 40, and 50 percent chord and with thickness ratios varying from 6 percent to 15 percent. [NACA 6 series Airfoil database search](#)  
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The NACA airfoil series The early NACA airfoil series, the 4-digit, 5-digit, and modified 4-/5-digit, were generated using analytical equations that describe the camber (curvature) of the mean-line (geometric centerline) of the airfoil section as well as the section's thickness distribution along the length of the airfoil. Later

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The NACA airfoils are airfoil shapes for aircraft wings developed by the National

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*NACA airfoil*

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Reinhard Subject:  $\frac{1}{2}$  $\frac{1}{2}$ The Naca Airfoil Series Clarkson University

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